

**Orbital Debris Assessment Report**

Prepared by ASTRA LLC  
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NASA DAS Software Version: 3.1.2

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Version 3

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consumption when the system is idle. RROCI has 6x 6000Ns/U Total Impulse and 480 g of total fuel. The fuel mass results in > 150 delV (m/s) and deorbit capability from > 800 km, which is far above RROCI orbit. The propulsion system will be used for intermediate attitude maintenance over the mission 1 year lifetime. At the end of the mission, ASTRA will initiate and control the full de-orbit and descent of the spacecraft. ASTRA does not expect any interaction or potential physical interference with other operational spacecraft, but can exercise defensive maneuvers if needed.

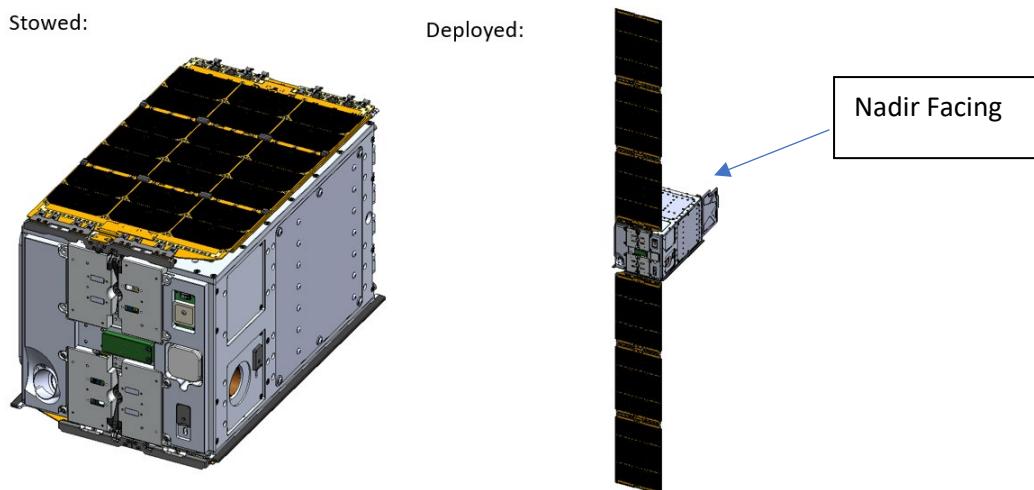
**Spacecraft On Orbit Attitude Control:** The RROCI spacecraft contains magnetometers, sun sensors, and star trackers that support an active, continuously engaged 3-axis attitude control system comprised of reaction wheels and torquer rods. The system specifications are shown below:

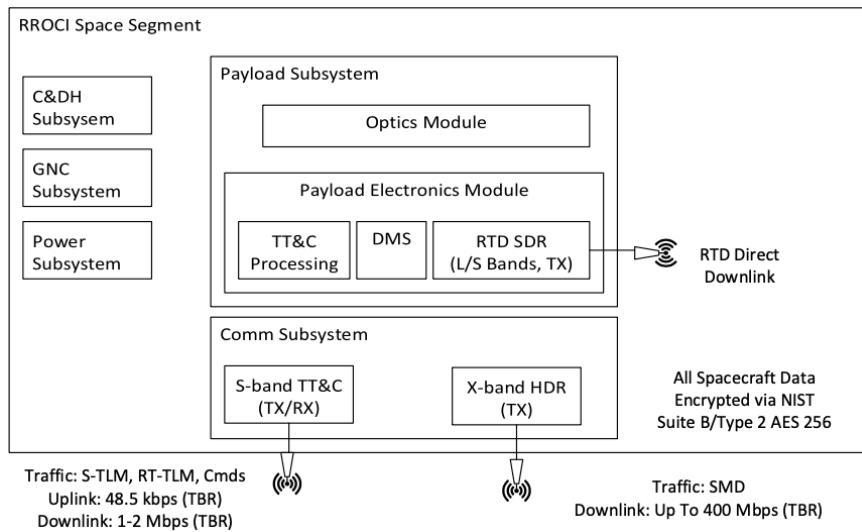
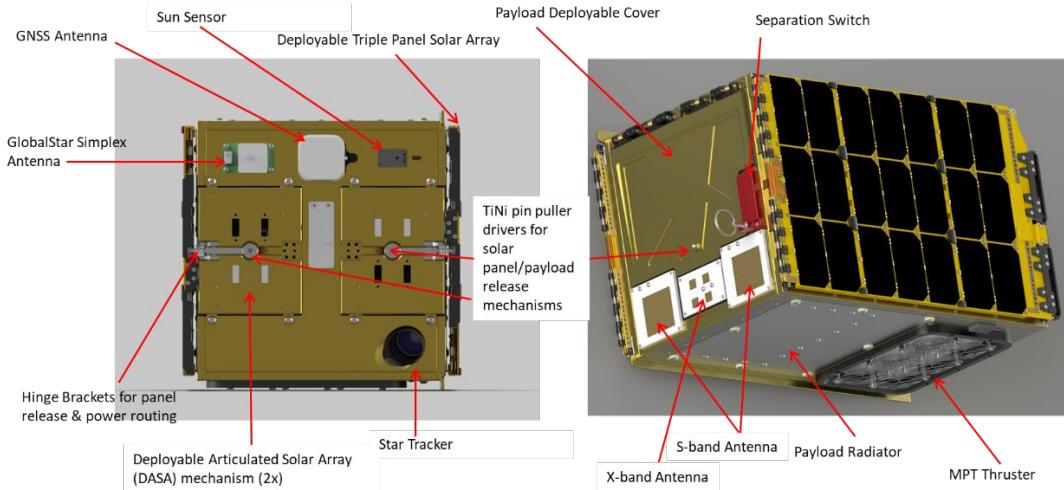
Performance		
Total momentum storage per axis	+/-15, +/-30, +/-50 <sup>7</sup>	mN.m.s
Maximum torque	2	mN.m
Magnetic moment	X/Y: 0.5, Z: 0.4	A.m <sup>2</sup>
Attitude determination accuracy	30	arcseconds
Pointing accuracy	<< 1	°
Slew rate	> 1.5 <sup>7</sup>	°/s
Radiation tolerance	> 45 <sup>8</sup>	krad (Si)
Operating temperature	- 45 / - 20 to + 40 / + 85 <sup>6</sup>	°C

**Spacecraft Power System:** The RROCI spacecraft uses solar panels to generate energy on-orbit, with local battery storage. There are no other sources of on-orbit energy generation or storage. The power design includes two triple panel wings (articulated) and two single panel wings (fixed) for a 128.0W peak power generation BoL and 115.8W peak power generation EoL. The batteries are 3500mAh 18650 Li-ion cells with 10A peak output current and 100Wh energy storage per battery, for a total of 168W peak power output.

## Section 2: Spacecraft Description

A detailed illustration of the RROCI spacecraft in the mission operation configuration with dimensional markings and marked component locations is shown below:





The system mass budget, at launch, is shown below:



(Requirement 4.5-1) Limiting Debris Generated by Collisions with Large Objects

Input				
Start Year	2022.3			
	Space	Perigee	Apogee	Inclination
	Structure	(km)	(km)	(deg)
RROCI	Payload	642	642	97.9

Output			
	Space	Compliance	Collision Probability
	Structure	Status	Probability
RROCI	Payload	Compliant	2.4873E-05

(Requirement 4.5-1) - Probability of Collision With Large Objects

The probability of collision with space objects, including orbital debris and meteoroids, of sufficient size to prevent postmission disposal is negligible. The RROCI propulsion system is redundant and comprised of solid propellant (i.e., a rupture of a gas tank is not applicable). There are no liquids on RROCI. The power system includes redundant solar panel strings and batteries. RROCI is compliant with requirement 4.5-2.

## **Section 6: Assessment of Spacecraft post-mission disposal plans and procedures**

At the end of the mission, ASTRA will initiate and control the full de-orbit and descent of the spacecraft using its on-board propulsion system (see Section 1). For RROCI, 6x 6000Ns/U Total Impulse and 480 g of total fuel. The fuel mass results in > 150 delV (m/s) and deorbit capability from > 800 km.

RROCI is fully compliant with Requirements 4.6-1 through 4.6-4 by utilizing the on-board propulsion for a controlled re-entry for postmission disposal below the 642km initial altitude utilizing a circular orbit altitude reduction approach as outlined in the NASA Spacecraft Conjunction Assessment and Collision Avoidance Best Practices Handbook (NASA/SP-20205011318) section 4.1, requiring no direct coordination with NASA HSF offices for ISS or Chinese Space Station collision avoidance.

## **Section 7: Assessment of Spacecraft Hazardous Materials**

- Detailed description of spacecraft components by size, mass, material, shape, and original location on the space vehicle, if the atmospheric reentry option is selected.

This information can be found in Appendix A, output from ODAR, with size, mass, material, and shape, with altitude of disintegration defined.

- Summary of objects expected to survive an uncontrolled reentry, using NASA DAS, NASA Object Reentry Survival Analysis Tool (ORSAT), or comparable software

This information can be found in Appendix A, output from ODAR, lines 72, 299, 304, 305.

- Calculation of probability of human casualty for the expected year of uncontrolled reentry and the spacecraft orbital inclination

Output from the ODAR analysis (NASA DAS 3.1.2) is 1:100000000.

- Assessment of spacecraft compliance with Requirement 4.7-1

Spacecraft is compliant with Requirement 4.7-1.

#### Section 7A: Assessment of Spacecraft Hazardous Materials

- Summary of the hazardous materials contained on the spacecraft using all columns and the format in paragraph 4.7.4.10.

The only hazardous materials on RROCI are the battery modules, which are shown to re-enter at 62.5 km with no human casualty risk.

347	710-01552 Battery Module 2	273	2	Aluminum 6061-T6	Box	0.502	0.089	0.093	0.044	62.5	0	0
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#### **Section 8: Assessment for Tether Missions**

Not applicable.

**See Appendix A for DAS Activity Log**















419	CABLE- SENSOR- TOP RADIATOR	1	1	Copper Alloy	Cylinder	0.0144	0.004	0.21		77.5	0	0
420	CONN- MICROSTRIP- SOCKET- 2POS- 26AWG- SOLDER CUP- CENTERLATCH 2	419	1	LCP Thermoplastic	Box	0.0001	0.003	0.013	0.003	77.5	0	0
421	SENSOR- TEMPERATURE 11	419	1	Copper Alloy	Cylinder	0.0059	0.015	0.008		76.4	0	0
422	WIRE- STRANDED- 30 AWG- PTFE- BLACK	419	1	Copper Alloy	Cylinder	0.0042	0.002	0.21		77.3	0	0
423	WIRE- STRANDED- 30 AWG- PTFE- WHITE 1	419	1	Copper Alloy	Cylinder	0.0042	0.002	0.21		77.3	0	0
424	CABLE- SENSOR- PEM RADIATOR	1	1	Copper Alloy	Cylinder	0.001	0.002	0.051		77.8	0	0
425	CONN- MICROSTRIP- SOCKET- 2POS- 26AWG- SOLDER CUP- CENTERLATCH 3	424	1	LCP Thermoplastic	Box	0.0001	0.003	0.013	0.003	77.8	0	0
426	Thermistor 2	424	1	Lead Element	Cylinder	0.0001	0.002	0.044		77.8	0	0
427	WIRE- STRANDED- 30 AWG- PTFE- WHITE 2	424	1	Copper Alloy	Cylinder	0.0008	0.002	0.051		77.6	0	0
428	CABLE- SENSOR- SC PAYLOAD CAVITY	1	1	Copper Alloy	Cylinder	0.001	0.002	0.051		77.8	0	0
429	CONN- MICROSTRIP SOCKET- 2POS- 26 AWG- SOLDER CUP- CENTER LATCH 3	428	1	LCP Thermoplastic	Box	0.0001	0.003	0.013	0.003	77.8	0	0
430	Thermistor 3	428	1	Lead Element	Cylinder	0.0001	0.002	0.044		77.8	0	0
431	WIRE- STRANDED- 30 AWG- PTFE- WHITE 3	428	1	Copper Alloy	Cylinder	0.0008	0.002	0.051		77.6	0	0